



## Airworthiness Directive

**AD No.:** 2025-0213

**Issued:** 26 September 2025

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I Part M.A.301, or Annex Vb Part ML.A.301, as applicable, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I Part M.A.303, or Annex Vb Part ML.A.303, as applicable] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

### Design Approval Holder's Name:

LEONARDO S.p.A.

### Type/Model designation(s):

AW189 helicopters

**Effective Date:** 10 October 2025

**TCDS Number(s):** EASA.R.510

**Foreign AD:** Not applicable

**Supersedure:** None

## ATA 71 – Power Plant – Engine Mount Fitting – Inspections / Replacement

### Manufacturer(s):

Leonardo S.p.A. Helicopters, formerly Finmeccanica S.p.A., AgustaWestland S.p.A.

### Applicability:

AW189 helicopters, serial numbers from 92001 to 92010 inclusive.

### Definitions:

For the purpose of this AD, the following definitions apply:

**The SB:** Leonardo Alert Service Bulletin (SB) 189-402.

**Affected part:** Engine external fitting right-hand (RH) Part Number (P/N) 8G5333A04351 and engine external fitting left-hand (LH) P/N 8G5333A04551.

**Serviceable part:** An affected part that is new (never previously installed on a helicopter).

### Reason:

Occurrences of heavy corrosion damage have been reported on LH and RH engine mount external fittings installed on AW189 helicopters.

This condition, if not detected and corrected, could reduce the structural integrity of the engine



mount.

To address this potential unsafe condition, Leonardo issued the SB providing instructions for repetitive inspections of the engine mount external fitting, and, depending on findings, replacement.

For the reasons described above, this AD requires accomplishment of repetitive inspections of the affected part, and, depending on findings, its replacement.

#### **Required Action(s) and Compliance Time(s):**

Required as indicated by this AD, unless the action(s) required by this AD have been already accomplished:

#### **Inspections:**

- (1) Within 100 flight hours (FH) after the effective date of this AD and, thereafter, at intervals not to exceed 400 FH, inspect each affected part in accordance with the instructions of Part I of the Accomplishment Instructions of the SB.
- (2) Within 24 months, or concurrently with the first engine 1 removal, whichever occurs first after the effective date of this AD, inspect the LH affected part in accordance with the instructions of Part II of the Accomplishment Instructions of the SB.
- (3) Within 24 months, or concurrently with the first engine 2 removal, whichever occurs first after the effective date of this AD, inspect the RH affected part in accordance with the instructions of Part II of the Accomplishment Instructions of the SB.

#### **Corrective Action(s):**

- (4) If, during any inspection as required by paragraph (1), (2) or (3) of this AD, as applicable, any crack or corrosion is detected on an affected part, before next flight, contact Leonardo Product Engineering Support for applicable instructions and, within the compliance time specified therein, accomplish those instructions accordingly.

#### **Terminating Action:**

- (5) Replacement of the affected part (LH or RH, respectively) with a serviceable part, accomplished on a helicopter in accordance with the instructions of the SB, constitutes terminating action for the repetitive inspections as required by paragraph (1) of this AD for the engine external fitting (LH or RH, respectively) of that helicopter.
- (6) If, during the inspection as required by paragraph (2) or (3) of this AD, as applicable, no crack or corrosion is detected on an affected part, that inspection constitutes terminating action for the repetitive inspections as required by paragraph (1) of this AD for that affected part.

#### **Ref. Publications:**

Leonardo SB 189-402 original issue dated 11 August 2025.

The use of later approved revisions of the above-mentioned document is acceptable for compliance with the requirements of this AD.



**Remarks:**

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. Based on the required actions and the compliance time, EASA have decided to issue a Final AD with Request for Comments, postponing the public consultation process until after publication. All interested persons may send their comments, referencing the AD Number, to the E-mail address specified in below Remark 3, prior to 24 October 2025. Only if any comment is received during the consultation period, a Comment Response Document will be published in the [EASA Safety Publications Tool](#), in a compressed ('zipped') file, attached to the record for this AD.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: [ADs@easa.europa.eu](mailto:ADs@easa.europa.eu).
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this AD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.
5. For any question concerning the technical content of the requirements in this AD, please contact: Leonardo S.p.A. Helicopters, E-mail: [engineering.support.lhd@leonardocompany.com](mailto:engineering.support.lhd@leonardocompany.com).

